

### Acknowledgment

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### References

- <sup>1</sup>Rumelhart, D., and McClelland, J., *Parallel Distributed Processing*, MIT Press, Cambridge, MA, 1986, pp. 318–330.
- <sup>2</sup>Narendra, K. S., and Partasarathy, K., "Identification and Control of Dynamical Systems Using Neural Networks," *IEEE Transactions on Neural Networks*, Vol. 1, No. 1, 1990, pp. 4–27.
- <sup>3</sup>Chen, C. L., and Nutter, R. S., "An Extended Back-Propagation Learning Algorithm by Using Heterogeneous Processing Units," *Proceedings of the International Joint Conference on Neural Network, IJCNN '92* (Baltimore, MD), Vol. 3, Inst. of Electrical and Electronics Engineers, 1992, pp. 988–993.
- <sup>4</sup>De Villiers, J., and Barnard, E., "Back-Propagation Neural Nets with One and Two Hidden Layers," *IEEE Transactions on Neural Networks*, Vol. 4, Feb. 1993, pp. 136–141.

## Frequency Weighting for the $H_\infty$ and $H_2$ Control Design of Flexible Structures

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### Introduction

THE  $H_\infty$  and the related  $H_2$  controller design methodologies allow for the design of control systems that meet tracking requirements and, at the same time, maintain the binding disturbance rejection properties. To achieve this, the problem should be appropriately defined in the quantitative terms. For example, the frequency shaping filters are used to define tracking requirements or disturbance rejection performance of the closed-loop system. These frequency-dependent weights filters are used only in the controller design stage. They add to the complexity of the problem because in the process of design the number of system equations varies and their parameters are modified. As stated by Voth et al. in Ref. 1, p. 55, "The selection of the controller gains and filters as well as the controller architecture is an iterative, and often tedious, process which relies heavily on the designers' experience." It is shown in this Note that this task is simplified in the case of flexible structure control. If a structure model is in the modal representation, then the addition of a filter is equivalent to the multiplication of each row of the plant input matrix by a constant. The  $i$ th constant is the filter gain at the  $i$ th natural frequency of the structure. In this way, each natural mode is weighted separately. This approach addresses the system performance at the mode level, which simplifies what otherwise may be an ad hoc and tedious process.

### Properties of Flexible Structures and Filters

We assume that a flexible structure is in the modal representation. Its transfer function  $G$  has the state space representation  $(A, B, C)$ ,

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with  $n$  degrees of freedom (or number of flexible structure modes),  $2n$  states,  $p$  inputs, and  $q$  outputs. Denote  $\omega_i$  the  $i$ th natural frequency and  $\zeta_i$  the  $i$ th modal damping,  $i = 1, \dots, n$ . We assume low damping ( $\zeta_i < 0.1$  for all modes) and distinct natural frequencies. In the modal representation, the system matrix  $A$  is block diagonal with  $2 \times 2$  blocks on the diagonal, and  $B$  and  $C$  are divided into  $2 \times p$  and  $q \times 2$  blocks (see Ref. 2, pp. 12–14):

$$A = \text{diag}(A_i), \quad B = \begin{bmatrix} B_1 \\ B_2 \\ \vdots \\ B_n \end{bmatrix}, \quad C = [C_1 \quad C_2 \quad \cdots \quad C_n] \quad (1)$$

$$A_i = \begin{bmatrix} -\zeta_i \omega_i & \omega_i \\ -\omega_i & -\zeta_i \omega_i \end{bmatrix}$$

where  $i = 1, \dots, n$ . Denote the transfer function of the  $i$ th mode as  $G_i = C_i(j\omega I - A_i)^{-1}B_i$ , then one obtains the following decomposition of transfer function in modal coordinates:

$$G(\omega) = \sum_{i=1}^n G_i(\omega) \quad (2a)$$

$$G(\omega_i) \cong G_i(\omega_i), \quad i = 1, \dots, n \quad (2b)$$

Equation (2a) can be derived by introducing  $A, B$ , and  $C$  as in Eq. (1) to the definition of the transfer function. Equation (2b) follows from Eq. (2a), by noting that the response at frequency  $\omega_i$  is dominated by the  $i$ th mode, that is,  $\|G_j(\omega_i)\|_2 \ll \|G_i(\omega_i)\|_2$  for  $i \neq j$ .

Consider a filter, with a diagonal transfer function  $F(\omega)$ . The diagonal input (output) filter represents lack of coupling between the inputs (or outputs). Denote  $\alpha_i$  the magnitude of the filter response at the  $i$ th natural frequency,  $\alpha_i = |F(\omega_i)|$ . The filter is smooth if the slope of its transfer function near the structural resonance is small when compared to the slope of the structure near the resonance. With these assumptions, the following property of the  $H_\infty$  norm of a structure with a filter is valid:

$$\|G\|_\infty \cong \max_{i \in [1, n]} (\|G_i\|_\infty) \quad (3a)$$

$$\|GF\|_\infty \cong \max_{i \in [1, n]} (\|G_i \alpha_i\|_\infty) \quad (3b)$$

To prove Eq. (3a), note that

$$\|G\|_\infty = \sup_{\omega} \bar{\sigma}(G(\omega)) \cong \max_{i \in [1, n]} \bar{\sigma}(G(\omega_i)) \cong \max_{i \in [1, n]} \bar{\sigma}(G_i(\omega_i)) \quad (4a)$$

Both approximations in Eq. (4a) hold because the resonance response of the  $i$ th mode dominates the structural response, as given in Eq. (2). However, because  $\|G_i\|_\infty \cong \bar{\sigma}(G_i(\omega_i))$ , therefore, Eq. (3a) is valid. To prove Eq. (3b), note that for the smooth  $F$  the transfer function  $GF$  preserves the properties of a flexible structure given by Eq. (2); thus,

$$\begin{aligned} \|GF\|_\infty &= \sup_{\omega} \bar{\sigma}[G(\omega)F(\omega)] \cong \max_{i \in [1, n]} \bar{\sigma}[G(\omega_i)F(\omega_i)] \\ &\cong \max_{i \in [1, n]} \bar{\sigma}(G_i(\omega_i)\alpha_i) \end{aligned} \quad (4b)$$

In the preceding approximation we took into consideration that  $\sigma_k(GF) = \sigma_k(G|F|)$ , which can be proven as follows:

$$\begin{aligned} \sigma_k^2(GF) &= \lambda_k(F^*G^*GF) = \lambda_k(FF^*G^*G) = \lambda_k(|F|^2G^*G) \\ &= \lambda_k(|F|G^*G|F|) = \sigma_k^2(G|F|) \end{aligned} \quad (4c)$$

Equation (3) says that the largest modal peak response of a lightly damped structure determines the worst-case response. A similar property holds for the 2-norm of a structure with a filter

$$\|G\|_2^2 \cong \sum_{i=1}^n \|G_i\|_2^2 \quad (5a)$$

$$\|GF\|_2^2 \cong \sum_{i=1}^n \|G_i \alpha_i\|_2^2 \quad (5b)$$

Equation (5) says that the rms response of a lightly damped structure is approximately an rms sum of the responses of each mode. Also, a norm of a smooth filter in series with a flexible structure is approximately equal to the norm of a structure scaled by the filter gains at natural frequencies. Note that a result similar to Eq. (5) holds for a structure with a filter at the output.

### Approximate Frequency Weighting

In controller design the inputs of the plant transfer function are separated into two groups: the exogenous input  $w$  (which includes commands and disturbances), and the actuator input  $u$ . The system outputs consists of the performance output  $z$ , at which performance is evaluated, and the sensed output  $y$ . The  $H_\infty$  ( $H_2$ ) control problem consists of determining the stabilizing controller transfer function  $K$  such that the  $H_\infty$  ( $H_2$ ) norm of the closed-loop transfer function  $G$ , from  $w$  to  $z$ , is minimized over all realizable controllers  $K$ .

Frequency weighting of the exogenous inputs and performance outputs is a standard approach in the  $H_\infty$  ( $H_2$ ) design to define the required closed-loop properties (see, for example, Ref. 3). In this case a plant is augmented with the input and/or output shaping filters, forming an augmented plant model. Consider, for example, input shaping. Denote  $F$  the transfer function of the input filter and assume that it is smooth. The transfer function from  $w$  to  $z$  with the input filter is  $GF$ . The inputs of  $G$  are shaped independently; therefore, the filter transfer function matrix is square and diagonal.

Introduce the transfer function

$$\hat{G} = \sum_{i=1}^n \hat{G}_i$$

where  $\hat{G}_i = C_i(j\omega I - A_i)^{-1} \hat{B}_i$  and  $\hat{B}_i = B_i \alpha_i$ , where  $\hat{G}_i$  is a transfer function  $G_i$  with the scaled input matrix  $B_i$ . We will show that the  $H_\infty$  norms of both transfer function are approximately equal:

$$\|GF\|_\infty \cong \|\hat{G}\|_\infty \quad (6)$$

One can prove it using Eq. (3b), obtaining

$$\|GF\|_\infty \cong \max_{i \in \{1,n\}} \|G_i \alpha_i\|_\infty = \max_{i \in \{1,n\}} \|\hat{G}_i\|_\infty \cong \|\hat{G}\|_\infty$$

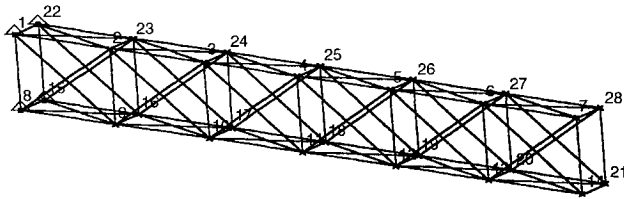


Fig. 1 Truss.

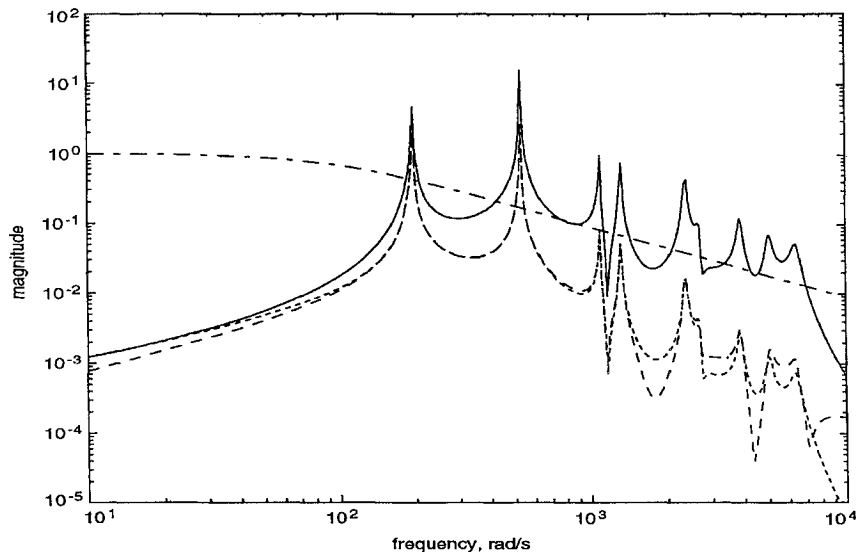


Fig. 2 Magnitudes of the open-loop transfer function: —, truss; ---, filter; ····, truss with filter; and - · - ·, truss with scaled disturbance input.

Equation (6) shows that the application of the input filter for the  $H_\infty$  performance modeling is equivalent to the scaling of the  $2 \times p$  input matrix  $B_i$  with  $\alpha_i$ , where  $\alpha_i$  is the magnitude of the filter transfer function at the resonant frequency  $\omega_i$ . For the  $H_\infty$  output filter one obtains

$$\|FG\|_\infty \cong \|\tilde{G}\|_\infty \quad (7)$$

where

$$\tilde{G} = \sum_{i=1}^n \tilde{G}_i, \quad \tilde{G}_i = \tilde{C}_i(j\omega I - A_i)^{-1} B_i, \quad \tilde{C}_i = \alpha_i C_i$$

For the  $H_2$  controller, the 2-norm of the transfer function  $GF$  is used as a system performance measure. We assume that the flexible structure is represented in the modal coordinates, and for the input filter we obtain

$$\|GF\|_2 \cong \|\hat{G}\|_2 \quad (8)$$

We prove it using Eq. (5), obtaining

$$\|GF\|_2^2 \cong \sum_{i=1}^n \|G_i \alpha_i\|_2^2 = \sum_{i=1}^n \|\hat{G}_i\|_2^2 = \|\hat{G}\|_2^2$$

Equation (8) shows that the application of the input filter for the  $H_2$  performance modeling is equivalent to the scaling of the  $2 \times p$  modal input matrix  $B_i$  with  $\alpha_i$ . For the  $H_2$  output filter we obtain

$$\|FG\|_2 \cong \|\tilde{G}\|_2 \quad (9)$$

Note that Eqs. (6–9) preserve the order and the physical interpretation of the transfer function and the corresponding state variables. This simplifies the controller design process because the relationship between the filter gains and the system performance is readily available.

### Example

Consider a steel truss as in Fig. 1. For this truss  $l_1 = 10$  cm,  $l_2 = 8$  cm, and the cross-sectional area is  $1 \text{ cm}^2$ . The disturbance  $w$  is applied at node 7 in the  $z$  direction, the performance  $z$  is measured at node 21 in the  $z$  direction, the input  $u$  is applied at node 20 in the  $z$  direction, and the output  $y$  is a displacement of node 28 in the  $z$  direction. The open-loop transfer function from the disturbance to the performance is shown in Fig. 2, as a solid line. The disturbance input is filtered with a low-pass filter,  $F(s) = 1/(1 + 0.011s)$ ; the magnitude of its transfer function is shown in the same figure, as a dot-dashed line. The resulting transfer function of the structure and

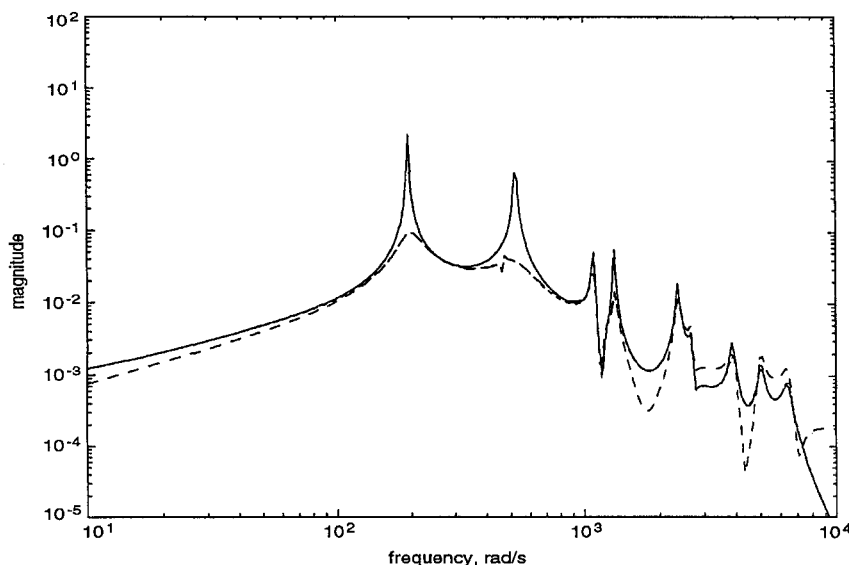


Fig. 3 Magnitudes of open-loop transfer function (—) and of the  $H_\infty$  closed-loop systems: ---, with the input shaping filter, and ···, with the scaled input matrix.

the filter is represented by the dotted line. The equivalent structure with the filter was obtained by scaling the disturbance input, according to Eq. (6), and the magnitude of its transfer function is shown in Fig. 2, as a dashed line. It is clear from that figure that the structure with the filter and the structure with the scaled disturbance input have very similar frequency characteristics, and their norms are  $\|G\|_\infty = 2.6903$  and  $\|G\|_2 = 453.2945$  for the structure with the filter and  $\|G\|_\infty = 2.6911$  and  $\|G\|_2 = 453.5661$  for the structure with the scaled disturbance input.

Two frequency weighted  $H_\infty$  controllers for this structure were designed. The first one is based on the structure with a filter, whereas the second is based on the structure with scaled input matrix. The open- and closed-loop transfer functions are shown in Fig. 3. The closed-loop performance of the structure with the filter and that with the scaled input is almost identical. The closed-loop  $H_\infty$  norms are as follows:  $\|G_{cl}\|_\infty = 0.09681$  for the structure with the filter and  $\|G_{cl}\|_\infty = 0.09676$  for the structure with the scaled disturbance input. In a similar manner,  $H_2$  controllers were designed. The closed-loop  $H_2$  norms are as follows:  $\|G_{cl}\|_2 = 108.6295$  for the structure with the filter, and  $\|G_{cl}\|_2 = 108.7181$  for the structure with the scaled disturbance input.

### Conclusions

It has been shown that for flexible structures the frequency shaping of the system properties with input (output) filters is equivalent to the scaling the modal input (output) matrix of the plant. This approach simplifies the controller design process. Instead of introducing new state variables, one modifies the gains of the modal input matrix. This is possible because the modal states related to the gains are weakly coupled, such that the modification of one state (or one gain) weakly influence the others. In addition, physical interpretation of the states remains unchanged and is related to the corresponding gains.

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### References

- <sup>1</sup>Voth, C. T., Richards, K. E., Jr., Schmitz, E., Gehling, R. N., and Morgenthaler, D. R., "Integrated Active and Passive Control Design Methodology for the LaRC CSI Evolutionary Model," NASA CR 4580, 1994.
- <sup>2</sup>Gawronski, W., *Balanced Control of Flexible Structures*, Springer, London, 1996.
- <sup>3</sup>Lim, K. B., and Balas, G. J., "Line-of-Sight Control of the CSI Evolutionary Model:  $\mu$  Control," *Proceedings of the American Control Conference*, Chicago, IL, 1992.

## Sliding Mode Controllers for Uncertain Systems with Input Nonlinearities

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### I. Introduction

THE stabilization problem of control systems with nonlinearities in the input has become a subject of keen interest in recent years, because nonlinearities inherently arise from practical actuators in system realization, for example, saturation, quantization, backlash, deadzone, and so on. The existence of nonlinear inputs is a source of degradation or, worse, shows instability in the performance of the system. Consequently, analysis of the problem of stability in control system design that accounts for nonlinearities in the input has become a subject of concern in research.<sup>1-5</sup> In addition to nonlinearities in the input, there are plant uncertainties, which originate from various sources, such as variation of plant parameters, inaccuracy arising from identification, etc. Therefore, the analysis of the problem of stability of robust control systems with plant uncertainties is as important as the problem of nonlinearities in the input.

In a robust control system, sliding mode control (SMC) is frequently adopted due to its inherent advantages of fast response and insensitivity to plant parameter variation and/or external perturbation. However, thus far, all of the published SMC studies concentrate on systems linear in the control input. The study of SMC for systems nonlinear in the control input has not been reported in the literature. In this study, a new SMC law is designed to ensure the global reaching condition of the sliding mode for uncertain systems with series nonlinearities. An example is given to verify the validity of the developed sliding mode controller.

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